



[4910-13-P]

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2012-1321; Directorate Identifier 2011-NM-147-AD]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede an existing airworthiness directive (AD) that applies to certain Airbus Model A310 series airplanes. The existing AD currently requires repetitive inspections for fatigue cracking of the area around the fasteners of the landing plate of the aileron access doors of the bottom skin panel of the wings, and related corrective action. The existing AD provides for an optional terminating action, which ends the repetitive inspections. Since we issued that AD, a reassessment of the previous fatigue threshold and inspection interval resulted in a determination that reduced inspection thresholds and intervals for accomplishment of the tasks are necessary. This proposed AD would reduce the initial inspection compliance time and intervals and provide additional terminating action options. We are proposing this AD to detect and correct fatigue cracking of the area around the fasteners of the landing plate of the aileron access doors and the bottom skin panel of the wings, which could result in reduced structural integrity of the wings.

DATES: We must receive comments on this proposed AD by [INSERT DATE 45

DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

ADDRESSES: You may send comments by any of the following methods:

- Federal eRulemaking Portal: Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- Fax: (202) 493-2251.
- Mail: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- Hand Delivery: U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC, between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Airbus SAS – EAW (Airworthiness Office), 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Operations office between 9 a.m. and 5 p.m., Monday through

Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone (800) 647-5527) is in the ADDRESSES section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the ADDRESSES section. Include “Docket No. FAA-2012-1321; Directorate Identifier 2011-NM-147-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

On July 9, 2004, we issued AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004). That AD required actions intended to address an unsafe condition on Airbus Model A310 series airplanes.

Since we issued AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004), the manufacturer has done a reassessment of the previous fatigue threshold and inspection interval resulting in a recommendation of reduced inspection thresholds and intervals for accomplishment of the tasks. The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA Airworthiness Directive 2011-0125, dated June 30, 2011 (referred to after this as “the MCAI”), to correct an unsafe condition for the specified products. The MCAI states:

DGAC [Direction Générale de l’Aviation Civile] France issued AD 2003-242(B) [which corresponds to FAA AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)] to require an inspection programme for aeroplanes with pre- and post- Airbus modification 05106 configurations (Airbus SB A310-57-2004) in order to detect any crack located on the trailing edge of the wing bottom skin No.2 panel of the all-speed-aileron servo control bay. A crack at this location, if not detected and corrected, would propagate towards the wing rear spar and ultimately into the wing fuel tank area. Undetected cracks would affect the structural integrity of the [left hand] LH and/or [right hand] RH wing.

Since issuance of DGAC France AD 2003-242(B), a reassessment of the previous fatigue threshold and inspection interval has been completed. As a result of the reassessment, the inspection thresholds and intervals for accomplishment of the tasks as defined in Airbus SB A310-57-2082 have been adjusted and reduced. Airbus SB

A310-57-2082 Revision 03 has been published, in which the compliance time periods for these inspection thresholds and intervals have been amended.

For the reasons stated above, this [EASA] AD retains the requirements of the DGAC France AD 2003-242(B), which is superseded, and requires implementation of the amended inspection programme.

Corrective action includes doing a permanent repair (installing a repair plate and new landing plates), a temporary repair (crack-stop drilling and application of a protective coating) followed by repetitive inspections until a permanent repair is done, and a repair approved by the FAA or EASA (or its delegated agent). This proposed AD also adds optional permanent repairs.

The initial inspection compliance times are dependent on the configuration (modification status, repair status, and crack length), and type of use (short range, long range, and normal). For airplanes without temporary repairs, the initial inspection compliance time ranges between 2,000 total flight cycles or 10,200 total flight hours, whichever occurs first; and 12,000 total flight cycles or 24,000 total flight hours, whichever occurs first. If the total flight cycles or total flight hours compliance time has been exceeded, the initial inspection compliance time (grace period) ranges between 200 flight cycles or 1,000 flight hours, to within 1,000 flight cycles or 2,800 flight hours, whichever occurs first.

For airplanes with temporary repairs, the initial inspection compliance time is dependant on crack length and ranges between 7 flight cycles or 35 flight hours, whichever occurs first, since the repair; to within 100 flight cycles or 200 flight hours, whichever occurs first, since the repair.

For airplanes with a temporary repair, the compliance time for completing the permanent repair ranges between 35 flight cycles or 175 flight hours, whichever occurs first, after completing the temporary repair; to within 500 flight cycles or 1,000 flight hours, whichever occurs first, after completing the temporary repair.

You may obtain further information by examining the MCAI in the AD docket.

Relevant Service Information

Airbus has issued the following service information.

- Airbus Service Bulletin A310-57-2081, Revision 03, dated October 13, 2010.
- Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010.
- Airbus Repair Instruction R573-49237, Repair To Crack In Trailing Edge Bottom Skin Panel No 2 Both Wings, Revision D, dated July 16, 2003.
- Airbus Repair Instruction R573-49243, Repair to cracks in all speed aileron access door landing plates, both wings, Revision C, dated July 16, 2003.

The actions described in this service information are intended to correct the unsafe condition identified in the MCAI.

FAA's Determination and Requirements of This Proposed AD

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information referenced above. We are proposing this AD because

we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products of the same type design.

Differences Between This AD and the MCAI or Service Information

Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010; and Airbus Service Bulletin A310-57-2081, Revision 03, dated October 13, 2010; specify to contact the manufacturer for instructions on how to repair certain conditions, but this proposed AD would require repairing those conditions using a method approved by the FAA or the EASA (or its delegated agent).

Costs of Compliance

Based on the service information, we estimate that this proposed AD would affect about 58 products of U.S. registry.

The actions that are required by AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004), and retained in this proposed AD take about 2 work-hours per product, at an average labor rate of \$85 per work hour. Based on these figures, the estimated cost of the currently required actions is \$170 per product.

We estimate that it would take about 4 work-hours per product to comply with the new basic requirements of this proposed AD. The average labor rate is \$85 per work-hour. Based on these figures, we estimate the cost of the proposed AD on U.S. operators to be \$19,720, or \$340 per product.

We have received no definitive data that would enable us to provide cost estimates for the on-condition actions specified in this proposed AD. We have no way of determining the number of products that may need these actions.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and

3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39 - AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by removing airworthiness directive (AD) 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004), and adding the following new AD:

Airbus: Docket No. FAA-2012-1321; Directorate Identifier 2011-NM-147-AD.

(a) Comments Due Date

We must receive comments by [INSERT DATE 45 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

(b) Affected ADs

This AD supersedes AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004).

(c) Applicability

This AD applies to Airbus Model A310-203, -204, -221, -222, -304, -322, -324, and -325 airplanes, certificated in any category, all serial numbers; except for airplanes identified in paragraphs (c)(1) and (c)(2) of this AD.

(1) Airplanes that have been modified in service according to Airbus Service Bulletin A310-57-2081 or during production by Airbus modification 12525.

(2) Airplanes that have been repaired according to Airbus Repair Inspection R573-49243 or R573-49237.

(d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

(e) Reason

This AD was prompted by a reassessment of the previous fatigue threshold and inspection interval specified in AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004), which resulted in a determination that reduced inspection thresholds and intervals for accomplishment of the tasks are necessary. We are issuing this AD to detect and correct fatigue cracking of the area around the fasteners of the landing plate of the aileron access doors and the bottom skin panel of the wings, which could result in reduced structural integrity of the wings.

(f) Compliance

You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

(g) Retained Repetitive Inspections for Airplanes Without Airbus Modification 5106

This paragraph restates the requirements of paragraph (a) of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004). For airplanes on which Airbus Modification 5106 (Airbus Service Bulletin A310-57-2004, Revision 2, dated March 5, 1990) has not been done as of August 31, 2004 (the effective date of AD 2004-15-07): Within 2,000 flight cycles after the August 31, 2004 (the effective date of AD 2004-15-07) , or within 3,000 flight cycles after the last inspection done per paragraph (k) of AD 98-26-01, Amendment 39-10942 (63 FR 69179, December 16, 1998), whichever is first; do a high frequency eddy current (HFEC) inspection for cracking of the area around the fasteners of the landing plate of the wing bottom skin panel No. 2 of the left and right wings. Do the inspection per the Accomplishment Instructions of Airbus Service Bulletin A310-57-2082, dated June 11, 2002. If no cracking is found, repeat the inspection thereafter at intervals not to exceed 1,900 flight cycles, until accomplishment of the terminating action specified in paragraph (j) of this AD. Accomplishment of the inspection required by paragraph (k) of this AD terminates the requirements of paragraph (g) of this AD.

(h) Retained Repetitive Inspection for Airplanes with Airbus Modification 5106

This paragraph restates the requirements of paragraph (b) of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004). For airplanes on which Airbus Modification 5106 has been done as of August 31, 2004 (the effective date of

AD 2004-15-07): Do the HFEC inspection required by paragraph (g) of this AD at the applicable time specified in paragraph (h)(1), (h)(2), (h)(3), or (h)(4) of this AD. If no cracking is found, repeat the inspection thereafter at intervals not to exceed 1,900 flight cycles, until accomplishment of the terminating action specified in paragraph (j) of this AD. Accomplishment of the inspection required by paragraph (k) of this AD terminates the requirements of paragraph (h) of this AD.

(1) For airplanes that have accumulated fewer than 17,000 total flight cycles since the date of issuance of the original Airworthiness Certificate or the date of issuance of the original Export Certificate of Airworthiness, whichever is first, as of August 31, 2004 (the effective date of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)): Inspect prior to the accumulation of 18,000 total flight cycles.

(2) For airplanes that have accumulated 17,000 or more total flight cycles, but fewer than 19,001 total flight cycles since the date of issuance of the original Airworthiness Certificate or the date of issuance of the original Export Certificate of Airworthiness, whichever is first, as of August 31, 2004 (the effective date of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)): Inspect within 2,000 flight cycles after August 31, 2004 (the effective date of AD 2004-15-07).

(3) For airplanes that have accumulated 19,001 or more total flight cycles, but fewer than 21,001 total flight cycles since the date of issuance of the original Airworthiness Certificate or the date of issuance of the original Export Certificate of Airworthiness, whichever is first, as of August 31, 2004 (the effective date of

AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)): Inspect with 1,200 flight cycles after August 31, 2004 (the effective date of AD 2004-15-07).

(4) For airplanes that have accumulated 21,001 or more total flight cycles since the date of issuance of the original Airworthiness Certificate or the date of issuance of the original Export Certificate of Airworthiness, whichever is first, as of August 31, 2004 (the effective date of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004)): Inspect within 500 flight cycles after August 31, 2004 (the effective date of AD 2004-15-07).

(i) Retained Corrective Action

This paragraph restates the requirements of paragraph (c) of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004). If any cracking is found during any inspection required by paragraph (g) or (h) of this AD: Before further flight, do the actions required by either paragraph (i)(1) or (i)(2) of this AD.

(1) Do a permanent repair of the area by doing the applicable corrective actions per the Accomplishment Instructions of Airbus Service Bulletin A310-57-2082, dated June 11, 2002. Accomplishment of the permanent repair terminates the repetitive inspections required by this AD for the repaired area only.

(2) Do the terminating action specified in paragraph (j) of this AD.

(j) Retained Optional Terminating Action, with New Service Information and Options

This paragraph restates the retained optional terminating action information specified in paragraph (d) of AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004), with new service information and new options. Modification of the landing plate of the aileron access doors of the wing bottom skin panel No. 2 of the left

and right wings by doing all the actions, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A310-57-2081, dated June 11, 2002, or Airbus Service Bulletin A310-57-2081, Revision 03, dated October 13, 2010; or by doing the repair in accordance with Airbus Repair Instruction R573-49243, Revision C, dated July 16, 2003, or Airbus Repair Instruction R573-49237, Revision D, dated July 16, 2003; terminates the requirements of this AD. Where Airbus Service Bulletin A310-57-2081, dated June 11, 2002; and Airbus Service Bulletin A310-57-2081, Revision 03, dated October 13, 2010; specify contacting the manufacturer for disposition of certain repair conditions that might be associated with the modification procedure, this AD requires that the repair be done in accordance with a method approved by either the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; the Direction Générale de l'Aviation Civile (DGAC) (or its delegated agent); or the European Aviation Safety Agency (EASA) (or its delegated agent).

(k) New Inspections, Related Investigative Actions, and Corrective Actions

Except as specified in paragraph (m)(1) of this AD, at the applicable time specified in Paragraph 1.E., "Compliance," of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010: Do a high frequency eddy current (HFEC) inspection to detect cracking of the area around the fasteners of the landing plate of the wing bottom skin panel No. 2 of the left and right wings; and do all applicable corrective actions; in accordance with the Accomplishment Instructions of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, except as required by paragraph (m)(2) of this AD. Do all applicable corrective actions before further flight. Repeat the inspection of the area around the fasteners of the

landing plate of the wing bottom skin panel number 2 of the left and right wings thereafter at the applicable intervals, including the compliance times for post temporary repair inspections, specified in Paragraph 1.E., “Compliance,” of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, except as specified in paragraph (m)(3) of this AD. The temporary repair of cracks, as identified in Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, does not constitute terminating action for the repetitive inspections required by this AD. Accomplishment of the inspection required by this paragraph terminates the requirements of paragraphs (g) and (h) of this AD. Doing the modification specified in paragraph (j) of this AD terminates the repetitive inspections required by this paragraph.

(l) New Permanent Repair

For airplanes on which the temporary repair as specified in Airbus Mandatory Service Bulletin A310-57-2082 has been done, within the applicable time specified in Paragraph 1.E., “Compliance,” of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010: Do the permanent repair, in accordance with the Accomplishment Instructions of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, except as provided by paragraph (m)(2) of this AD.

(m) New Exceptions to Service Information

(1) Where Paragraph 1.E., “Compliance,” of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, specifies a compliance time “from receipt of this service bulletin,” this AD requires compliance within the specified compliance time after the effective date of this AD.

(2) Where Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, specifies to contact Airbus for repair: Before further flight, repair the crack using a method approved by either the Manager, International Branch, ANM-116; or EASA (or its delegated agent).

(3) Where Paragraph 1.E., “Compliance,” of Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010, specifies to contact Airbus for inspection intervals, this AD requires using an inspection interval approved by either the Manager, International Branch, ANM-116; or EASA (or its delegated agent).

(n) Credit for Previous Actions

(1) This paragraph provides credit for the actions required by paragraphs (k) and (l) of this AD, if those actions were performed before the effective date of this AD using the service information specified in paragraph (n)(1)(i), (n)(1)(ii), or (n)(1)(iii) of this AD, which is not incorporated by reference in this AD.

(i) Airbus Service Bulletin A310-57-2082, dated June 11, 2002.

(ii) Airbus Service Bulletin A310-57-2082, Revision 01, dated August 22, 2003.

(iii) Airbus Service Bulletin A310-57-2082, dated June 11, 2002.

(2) This paragraph provides credit for the modification of the landing plate of the aileron access doors of the wing bottom skin panel No. 2 of the left and right wings required by paragraph (j) of this AD, if those actions were performed before the effective date of this AD using the service information specified in paragraph (n)(2)(i) or (n)(2)(ii) of this AD, which is not incorporated by reference in this AD, except where this service information specifies contacting the manufacturer for disposition of certain repair conditions that might be associated with the modification procedure, this AD requires

that the repair be done in accordance with a method approved by either the Manager, International Branch, ANM-116; or the EASA (or its delegated agent).

(i) Airbus Service Bulletin A310-57-2081, Revision 01, dated February 26, 2003.

(ii) Airbus Service Bulletin A310-57-2081, Revision 02, dated October 18, 2007.

(o) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM-116, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone (425) 227-2125; fax (425) 227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Airworthy Product: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is

airworthy before it is returned to service.

(3) AMOCs approved previously in accordance with AD 2004-15-07, Amendment 39-13741 (69 FR 44592, July 27, 2004), are approved as AMOCs for the corresponding provisions of this AD.

(p) Related Information

(1) Refer to MCAI EASA Airworthiness Directive 2011-0125, dated June 30, 2011, and the service information specified in paragraphs (p)(1)(i) through (p)(1)(vi) of this AD, for related information.

(i) Airbus Mandatory Service Bulletin A310-57-2082, Revision 03, dated November 15, 2010.

(ii) Airbus Service Bulletin A310-57-2081, dated June 11, 2002.

(iii) Airbus Service Bulletin A310-57-2081, Revision 03, dated October 13, 2010.

(iv) Airbus Service Bulletin A310-57-2082, dated June 11, 2002.

(v) Airbus Repair Instruction R573-49237, Revision D, dated July 16, 2003.

(vi) Airbus Repair Instruction R573-49243, Revision C, dated July 16, 2003.

(2) For service information identified in this AD, contact Airbus SAS – EAW (Airworthiness Office), 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email

account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may review copies of the referenced service information at the FAA, Transport Airplane Directorate,

1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

Issued in Renton, Washington, on January 20, 2013.

Michael Kaszycki,
Acting Manager,
Transport Airplane Directorate,
Aircraft Certification Service.

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